

Probabilistic Design of Hollow Airfoil Composite Structure by Using Finite Element Method

Mr. Sachin M. Shinde*, Dr .B. P. Ronge**, Dr. P. M. Pawar**

*(Assistant Professor, Department of Mechanical Engineering, Karmayogi Engineering College, Pandharpur)

** (Professor, Department of Mechanical Engineering, SVRI's College of Engineering, Pandharpur)

ABSTRACT

This study represents simulation of Airfoil composite beam by using Monte Carlo method. A three dimensional static analysis of large displacement type has been carried out. Finite element analysis of NACA0012 airfoil composite structure has been carried out and uncertainty in bending stress is analyzed. Bending stress was objective function. Chord length , beam length ,elastic modulus in XY,YZ,XZ and shear modulus of epoxy graphite in XY,YZ,XZ, ply angle and ply thickness of airfoil section, force are varied within effective range and their effect on bending stress has been analyzed. In order to validate the results, one loop of simulation is benchmarked from results in literature. Ultimately, best set of probabilistic design variable is proposed to reduce bending stress under static loading condition.

Key words: -Hollow Airfoil composite structure, Monte Carlo Simulation.

I. INTRODUCTION

Composite materials have found increasing use in aerospace and civil engineering construction. One of the common areas of application is panels and airfoils construction where composite materials with complex lay-ups are used. The following properties can be improved when composite materials are used: specific strength, specific stiffness, weight, and fatigue life. The thin-walled beams of open cross-sections are used extensively in space systems as space erectable booms installed on spacecraft; in aeronautical industry both as direct load-carrying members and as stiffener members. In addition, they are used as well in marine and civil engineering, whereas the I-beams, in the fabrication of flex beams of bearing less helicopter rotor [1]. Thin-walled structures are integral part of an aircraft [2]. That is the reason why many researchers consider it in their studies and published it in scholarly articles. Chan and his students focused on thin-walled beams with different cross-sections. Among their studies, Chan and Dermirhan [3] considered first a circular cross section thin walled composite beam. They developed a new and simple closed-form method to calculate its bending stiffness. Then, Lin and Chan [4] continued the work with an elliptical cross section thin-walled composite beam. Later, Syed and Chan [5] included hat-sectioned composite beams. And most recently, Rao and Chan [6] expanded the work to consider laminated tapered tubes. Ascione et al. [7] presented

a method that formulates a one-dimensional kinematical model that is able to study the static behavior of fiber-reinforced polymer thin-walled beams. It's well known that the statics of composite beam is strongly influenced by shear deformability because of the low values of the elastic shear module. Such a feature cannot be analyzed by Vlasov's theory , which assumes that the shear strains are negligible along the middle line of the cross-section. Ferrero et al. [8] proposed that the stress field in thin-walled composite beams due to at twisting moment is not correctly modeled by classical analytical theories, so numerical modeling is essential. Therefore, they developed a method with a simple way of determining stress and stiffness in this type of structures where the constrained warping effect can be taken into account. They worked with both open and closed cross sections. Also, to check the validity of the method for structures made of composite materials, a beam with thin, composite walls were studied. Wu et al. [9] presented a procedure for analyzing the mechanical behavior of laminated thin-walled composite box beam under torsional load without external restraint. Some analyses have been formulated to analyzed composite box beam with varying levels of assumptions [10-13]. Therefore, analysis of airfoil wing under varying loading condition is key to improve the design and provide good agreement in results.

II. SIMULATION

The Monte Carlo Simulation method is the most common and traditional method for a probabilistic analysis. This method simulates how virtual components behave the way they are built. Present work uses FEM package ANSYS for analysis of composite beam of hollow NACA0012 airfoil shape. Element selected for meshing the geometry of the specimen is shell 181. Material properties of epoxy graphite are entered. Geometry of model is drawn in ANSYS software. Geometry is meshed by giving element size 1 mm. Mapped type of meshing is used. Meshed model of specimen is shown below in figure in 1.

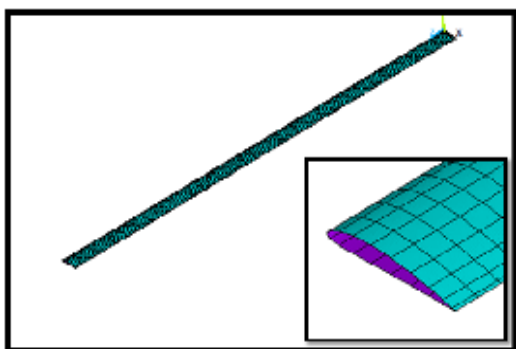


Figure 1: Meshed model of wing with SHELL 181 elements (zoomed cross section in box)

Meshed model contains 3549 number of nodes and 3360 number of elements. The mesh size is reasonably small to obtain fairly accurate results. Figure 2 shows model with applied loads and boundary conditions. Geometry is meshed with element size 5 mm. Mapped type of meshing is used. Meshed model of specimen is shown in above figure

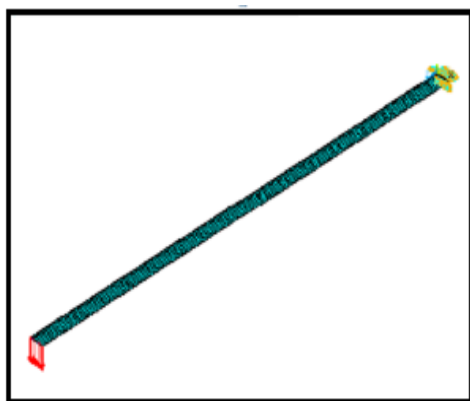


Figure 2: Meshed geometry with boundary conditions

III .RESULTS AND DISCUSSION

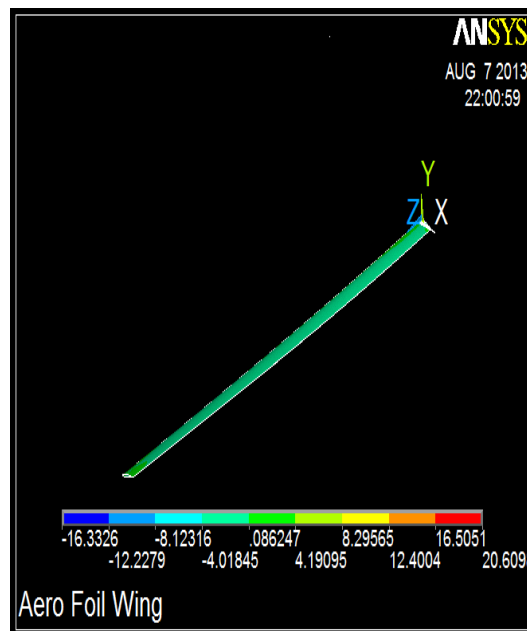


Figure 3: Contour plot of Bending stress distribution

Figure 3 shows bending stress distribution in composite airfoil beam. Scatter plot is obtained at end of static analysis. Maximum value of bending stress is 20.609 N/mm^2 and it is observed in the region at the end of beam. One loop of simulation is validated from results in literature.

Table1: Comparison of Literature and ANSYS results [16]

Hollow Airfoil composite beam	Bending stress (N/mm^2)		
	Literature	Current study	% Error
	18.93 N	20.609	11%

Probabilistic design system is used to determine the effect of one or more variables on the outcome of the Hollow Aerofoil composite beam. Present work considers:

- Geometric parameters: Length, Chord length.
- Material parameters: Young modulus, Poission ratio and Shear modulus in respective direction.
- Composite properties: Layer thickness and Orientation angles.

- Load parameters: Tip load.

Table 2: Parameters used in probabilistic design of NACA0012 hollow aerofoil beam [16]

Parameter	Distribution Type	Mean (or Baseline)	Standard Deviation
Length	Normal	1680 mm	16.80 mm
Chord length	Normal	56.0 mm	2.80 mm
EXX	Normal	142. x 10 ³ MPa	14200 MPa
EYY=EZ Z	Normal	9.80 x 10 ³ MPa	980 MPa
PRXY=P RXZ	Normal	0.42 MPa	0.084 MPa
PRYZ	Normal	0.50 MPa	0.10 MPa
GXY=G XZ	Normal	6 x 10 ³ MPa	600 MPa
GYZ	Normal	4 x 10 ³ MPa	400 MPa
Layer Thickness	Normal	0.127 mm	0.0127 mm
θ	Normal	60 deg.	6.0 deg.
Force	Normal	-1.0 N	0.1 N

All the parameters are considered as varying with Gaussian (or Normal) distribution (see Table 2). Using uncertainties as stated above, probabilistic design system is performed using ANSYS to know sensitivity of each parameter on Bending Stress. PDS within ANSYS uses Monte Carlo Simulation approach and analysis was looped through 1000 sample points considering the variations defined in the input variables and the corresponding static analysis of the output parameters.

This section presents results of probabilistic design system for Aerofoil composite beam. An overview of the data is provided by several plots from Figures 4 to 17. Figures 4 to 17 show Scatter plots each input and output parameters. Scatter plots show uncertainty in Bending Stress. Polynomial distribution of C1 powers is indicated by red colored line. As degree of polynomial distribution is small, there is more uncertainty in Bending Stress. If linear correlation coefficient of scatter plot is small then there is less uncertainty in Bending Stress and if larger

then there is more uncertainty in Bending Stress. The same is true for rank order correlation coefficient.

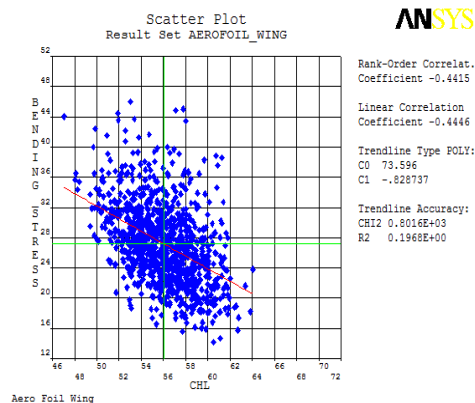


Figure 4: Scatter plot of Bending Stress vs Chord Length of hollow aerofoil composite beam

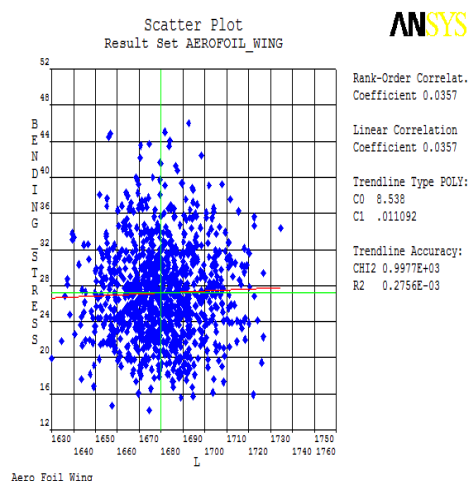


Figure 5: Scatter plot of Bending Stress vs Beam Length of hollow aerofoil composite beam

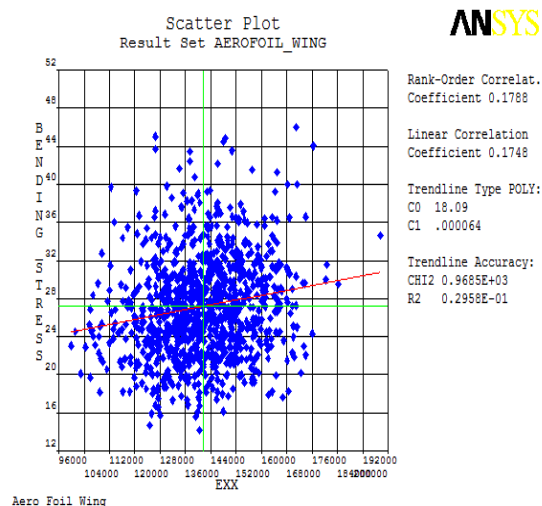


Figure 6: Scatter plot of Bending Stress vs EXX of hollow aerofoil composite beam

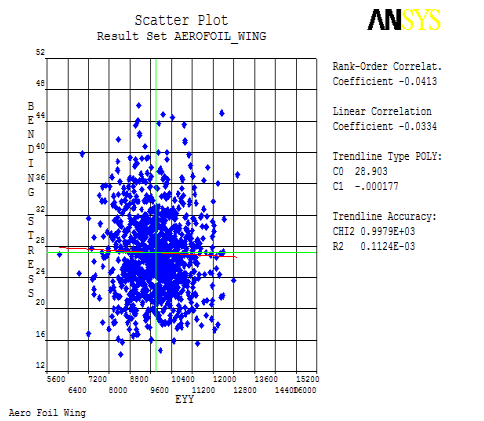


Figure 7: Scatter plot of Bending Stress vs EYY of hollow aerofoil composite beam

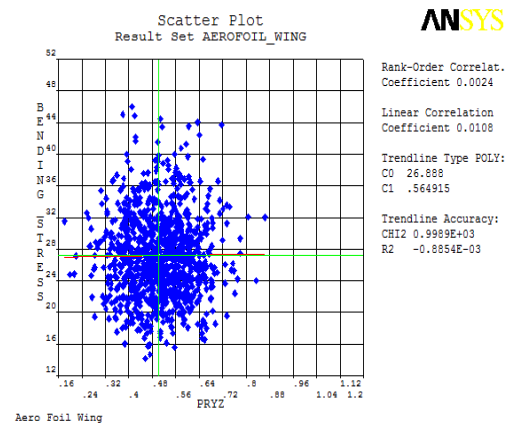


Figure 10: Scatter plot of Bending Stress vs PRYZ of hollow aerofoil composite beam

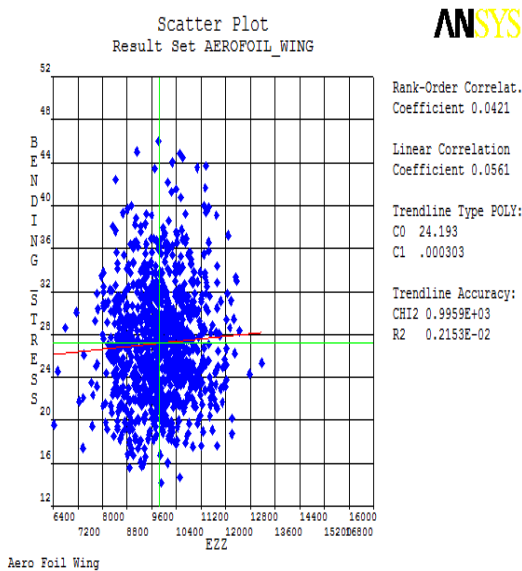


Figure 8: Scatter plot of Bending Stress vs EZZ of hollow aerofoil composite beam

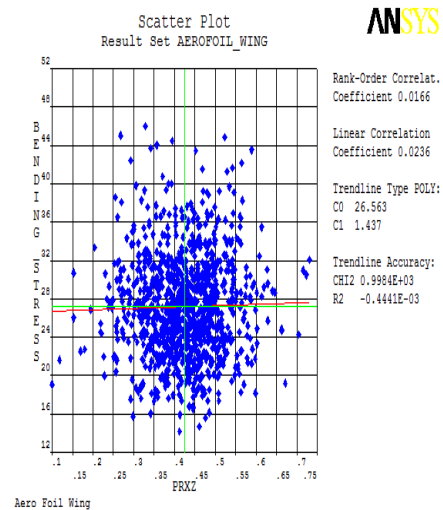


Figure 11: Scatter plot of Bending Stress vs PRXZ of hollow aerofoil composite beam

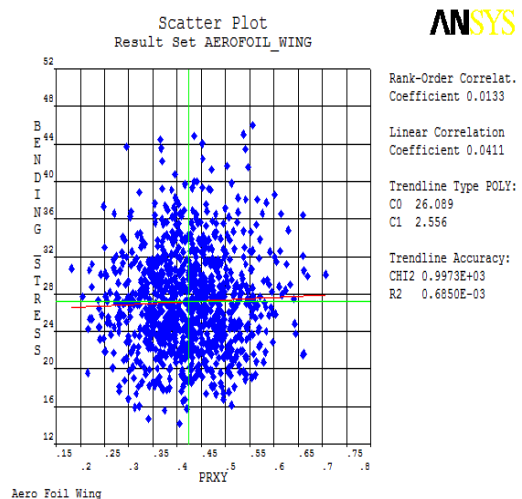


Figure 9: Scatter plot of Bending Stress vs PRXY of hollow aerofoil composite beam

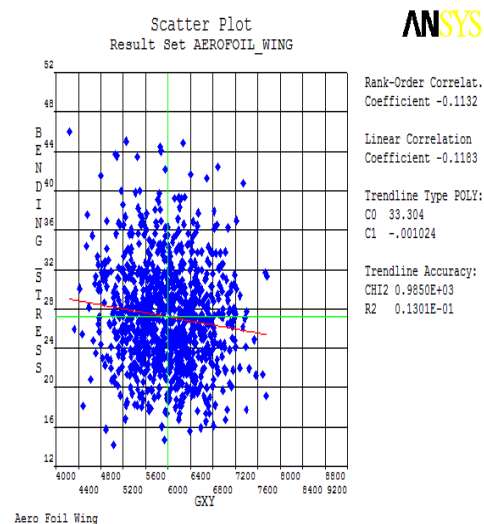


Figure 12: Scatter plot of Bending Stress vs GXY of hollow aerofoil composite beam

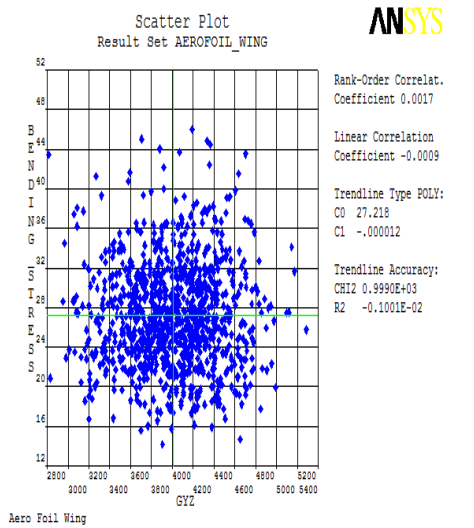


Figure 13: Scatter plot of Bending Stress vs GYZ of hollow aerofoil composite beam

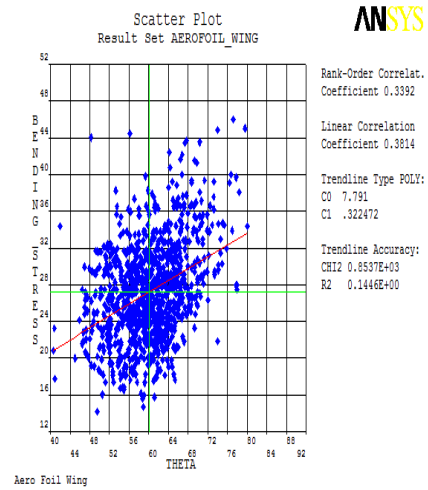


Figure 16: Scatter plot of Bending Stress vs Ply Angle (THETA) of hollow aerofoil composite beam

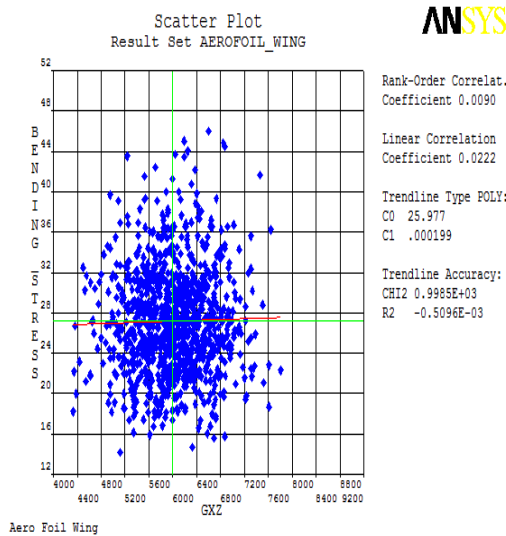


Figure 14: Scatter plot of Bending Stress vs GXZ of hollow aerofoil composite beam

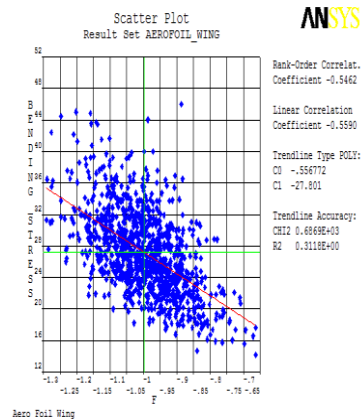


Figure 17: Scatter plot of Bending Stress vs Load of hollow aerofoil composite beam

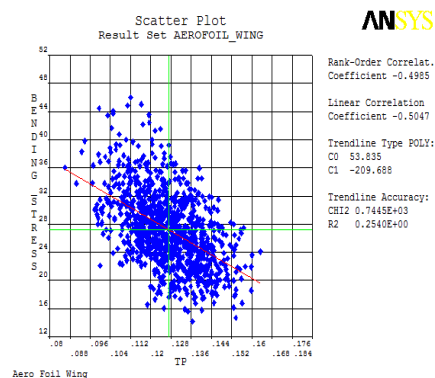


Figure 15: Scatter plot of Bending Stress vs Ply Thickness of hollow aerofoil composite beam

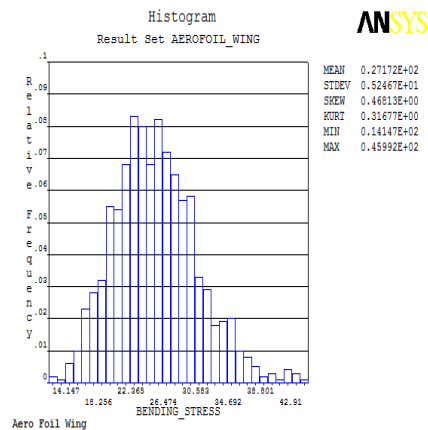


Figure 18: Histogram of output parameter Bending Stress of hollow aerofoil composite beam

Figure 18 shows variation in Bending Stress due to combined variation in various input parameters. Bending Stress varies between 14.14 MPa to 45.99 MPa. Although all input parameters vary using normal distribution function but output parameters do not follow same. It can be seen from values of Kurtosis and Skewness. Value of skewness deviates from zero.

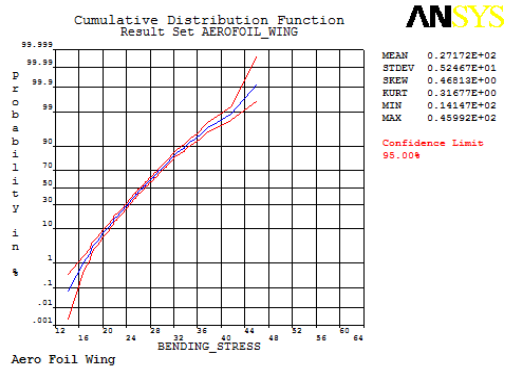


Figure 19: 95% confidence interval for Bending Stress of hollow aerofoil composite beam

Beams are typically designed to fulfill certain design criteria based on the output parameters. For example, a design for Bending Stress is that the deflection will be above or below a certain limit. The cumulative distribution curve for Bending Stress shows minimum 14.14 MPa and maximum 45.99 MPa. Probability of having Bending Stress less than 27 MPa is below 50% (see Figure 19).

The line in middle is the probability P that the maximum Bending Stress remains lower than a certain limit value with 95% confidence interval. The confidence interval quantifies the accuracy of the probability results. After the reliability of the beam has been quantified, it may happen that the resulting value is not sufficient. The answer to the question which input variables should be addressed to achieve a robust design and improve the quality; can be derived from probabilistic sensitivity diagrams plot.

Variation in following six parameters significantly affects Bending Stress of Aerofoil composite beam whereas variation in remaining eight parameters does not have significant effect:

- Tip Load
- Ply Thickness
- Chord Length
- Orientation angle
- Young modulus in XX
- Shear modulus in XY

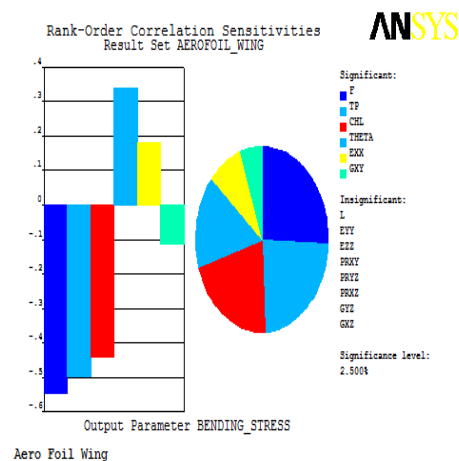


Figure 20: Sensitivity plot for Bending Stress of hollow areofoil composite beam for static analysis

IV. CONCLUSION

The influence of the design parameters on bending stress under variable loading condition is studied. The conclusions obtained are summarised as follows.

- Baseline analysis deflection results perfectly match with literature results for all three cases and percentage error is less than 3%.
- Successfully carried out probabilistic analysis to study effect of input uncertainties on Bending Stress of static analysis for all three composite beams. From analysis it appears that not all input uncertainties affect Bending stress. This analysis is also used to determine limits for various inputs during optimization

REFERENCES

- [1]. Lee, J., and Lee, S., "Flexural-torsional behavior of thin-walled composite beams", ELSEVIER, Thin-walled Structures, vol. 42, 2004, pp. 1293-1305.
- [2]. Mitra, M., Gopalakrishnan, S., and Seetharama, M., "A new super convergent thin walled composite beam element for analysis of box beam structures", ELSEVIER, International Journal of Solids and Structures, vol. 41, 2004, pp. 1491-1518.
- [3]. Chan, W. S., and Demirhan K. C., "A Simple Closed-Form Solution of Bending Stiffness for Laminated Composite Tubes", Journal of Reinforced Plastic & Composites, vol. 19, 2000, pp. 278-291.
- [4]. Lin, C. Y., and Chan, W. S., "A Simple Analytical Method for Analyzing Laminated

- Composites Elliptical Tubes”, Proceedings of the 17th Technical Conference of American Society of Composites.
- [5]. Syed, K. A., and Chan, W. S., “Analysis of Hat-Sectioned Reinforced Composite Beams”, Proceedings of American Society for Composites Conference.
- [6]. Rao, C., and Chan S., “Analysis of Laminated Composites Tapered Tubes”, Department of Mechanical and Aerospace Engineering, University of Texas at Arlington.
- [7]. Ascione, L., Feo, L., and Mancusi, “On the of fiber-reinforced polymer thin-walled beams”, ELSEVIER, Composites, Part B, vol. 31, 2000, pp. 643-654.
- [8]. Ferrero, J. F. , Barrau, J. J. , Segura, J. M. , and Sudre, M., “Torsion of thin-walled composite beams with mid plane symmetry”, ELSEVIER, Composite Structures, vol. 54, 2001, pp. 111-120.251
- [9]. Wu, Y., Zhu, Y., Lai, Y., Zhang, X., , “Analysis of thin-walled composite box beam under Torsional load without external restraint”, ELSEVIER, Thin-walled Structures, vol. 40, 2002, pp. 385-397.
- [10]. Chuanxian, C., “Researches on bending and torsional stiffness of thin-walled carbon epoxy box beam”, Mechanics and Practice, Beijing University Press, 1985.
- [11]. Chandra, R., Stemple, A. D., and Chopra I., “Thin-walled composite beams, torsional and extensional loads”, Journal Aircraft, 1990, vol. 27, 619-626.
- [12]. Fei, Y., “A theory for bending and torsion of the composite single cell thin-walled beam”, Mechanics and Practice, 1994, vol. 16, pp. 37-40.
- [13]. Min, J. S., Hyo, C. M., and In, L., “Static and dynamic analysis of composite box beams using large deflection theory”, Computer & Structures, 1995, vol. 57, pp. 635-642.
- [14]. Shinde, S.M., Ronge, B, P, Pawar, P. P., “Probabilistic Design and Random Optimization of Aerofoil Composite by Using Finite Element Method”, www.ijera.com Vol. 3, Issue 5, Sep-Oct 2013, pp.01-05
- [15]. Shinde S.M., “Probabilistic Design and Random Optimization of Circular Composite by Using Finite Element Method”, IJMET, Volume 5, Issue 4, April (2014), pp. 206-215
- [16]. Zhao, Y., HU Haiyan, “Structural Modeling and Aero elastic Analysis of High Aspect Ratio Composite Wings”, Institute of Vibration Engineering Research, Nanjing University of Aeronautics and Astronautics, Nanjing, 2005.